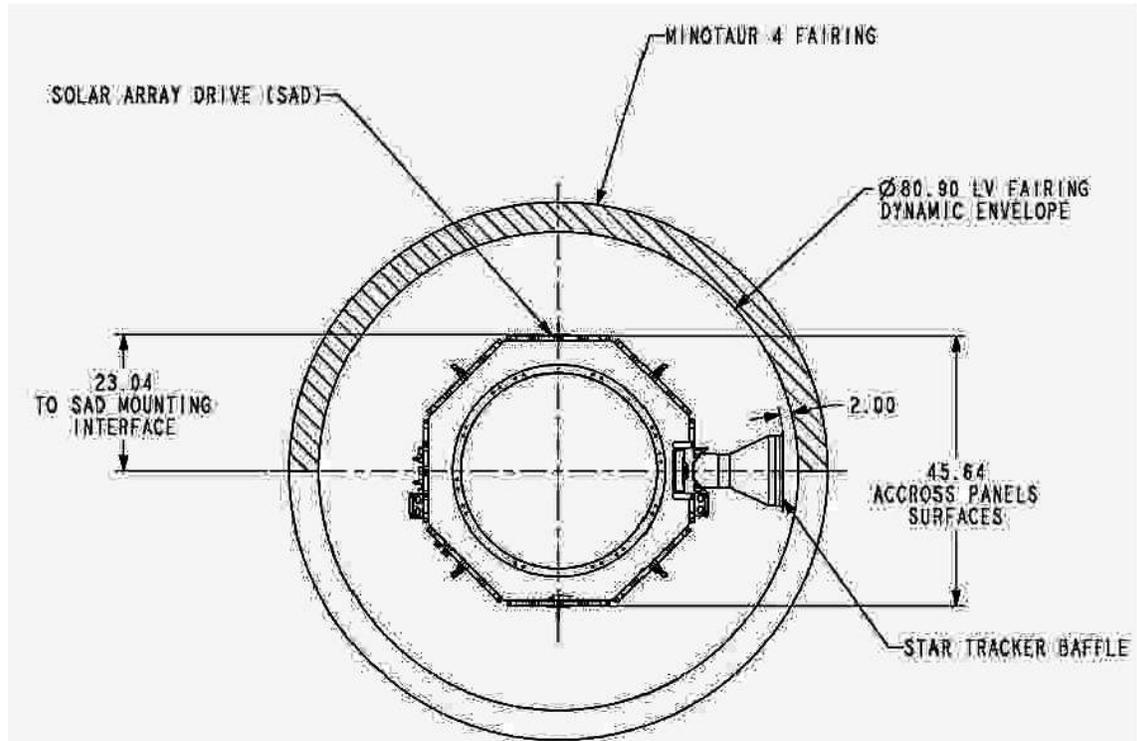


The purpose of this amendment is to make clarifications and changes to the specifications. The period in which proposals will be accepted is not extended.

Deleted texts are marked with ~~striketrough~~ and clarifications are marked in **bold**.

Paragraph 3.2.2, Figure 2 is deleted and replaced with the following:



Paragraph 4.2.1 is revised as follows:

#### 4.2.1 Temperature

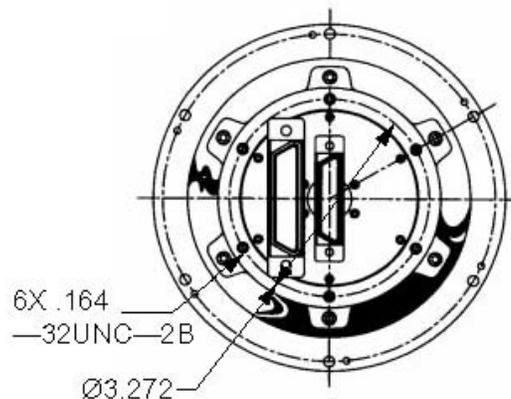
The arrays shall operate properly when the surroundings and the surface on which it is mounted are maintained at any temperature between  $-150^{\circ}\text{C}$  and  $+105^{\circ}\text{C}$   **$+120^{\circ}\text{C}$**  for prototype testing and on-orbit operations in the deployed configurations. During the orbital design lifetime the solar array will experience approximately 3000 thermal cycles.

Paragraph 5.1.2 is revised as follows with the addition of Figure 5:

### 5.1.2 SADA Interface - Deployed Configuration

The SADA mounting interface is ~~TBD~~ **shown in Figure 5**. The flight limit load applied to the S/C interface shall not exceed those specified below during its complete life.

- The maximum allowable combined bending moment is:  $M = 250 \text{ Nm}$  where  $M = \sqrt{(M_x^2 + M_z^2)}$
- The maximum allowable combined force is:  $F = 10,000 \text{ N}$  where  $F = \sqrt{(F_x^2 + F_y^2 + F_z^2)}$



**Figure 5: SADA Mounting Interface**

Paragraph 5.3.1 is revised as follows:

### 5.3.1 Solar Array Drive Assembly (SADA)

The SADA slip ring capabilities are listed in Table 8. The solar array wing shall mate to the SADA with a TBD connector. The SADA will have the capability to pass through a minimum of four (4) thermistors, ~~two (2) coarse sun sensors~~, and 1 (1) heater circuit per wing.

Paragraph 7.2.5.5.1 is revised as follows:

### 7.2.5.5.1 Protoflight Thermal Vacuum Cycling Limits

Protoflight level tests shall consist of N cycles to a maximum temperature of ~~+105°C~~ **+120°C** and a minimum temperature of -150°C as shown in Figure ~~5~~ **6**. A dwell time of 2 hours shall occur at each maximum and minimum temperature plateaus before continuing to the next cycle.

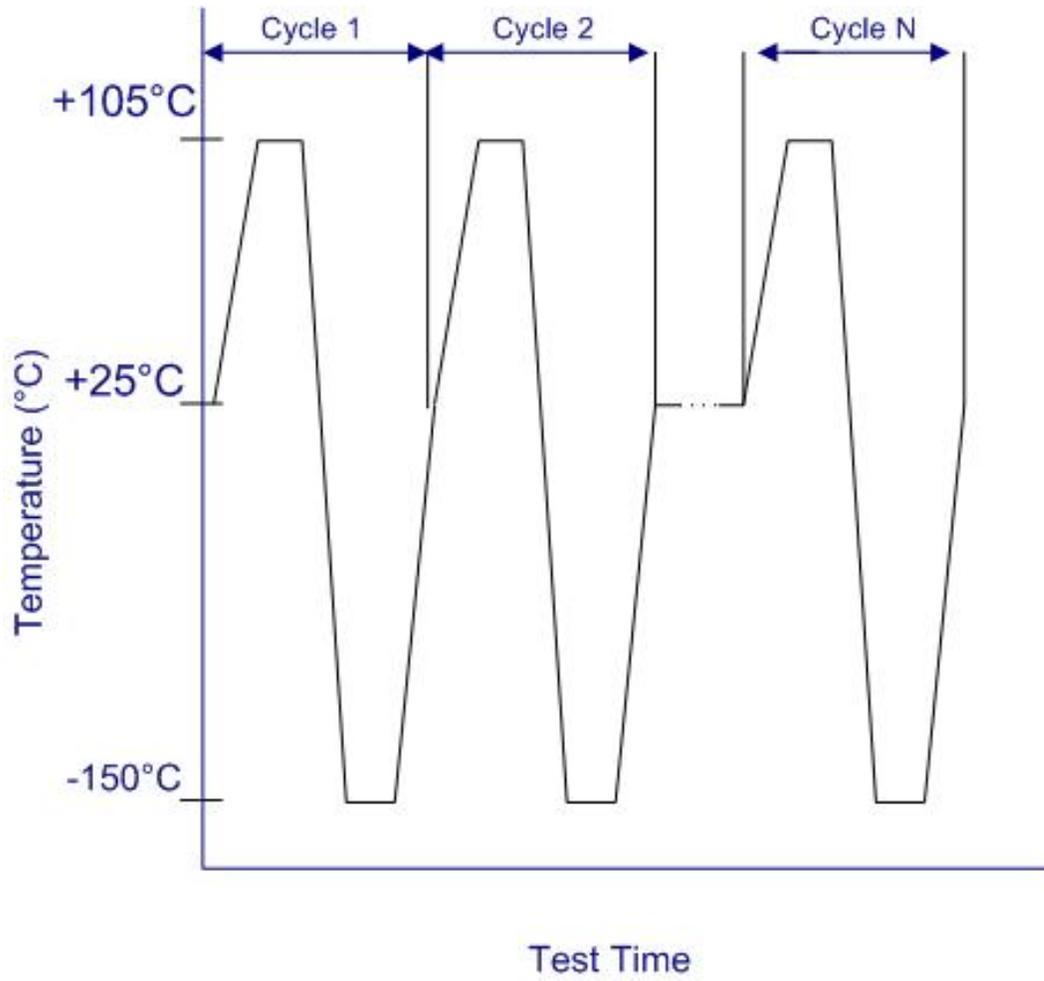


Figure 5 6: Protoflight Thermal Vacuum Profile